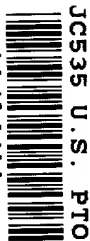


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UTILITY PATENT APPLICATION TRANSMITTAL (Only for new nonprovisional applications under 37 C.F.R. § 1.53(b))	Attorney Docket No.	13DVI3092
	First Inventor or Application Identifier	Sridhar Adibhatla
	Title	Methods and Apparatus for Model Based Diagnostics
	Express Mail Label No.	

APPLICATION ELEMENTS <i>See MPEP chapter 600 concerning utility patent application contents.</i>	ADDRESS TO: Assistant Commissioner for Patents Box Patent Application Washington, DC 20231	
1. <input checked="" type="checkbox"/> Fee Transmittal Form (e.g., PTO/SB/17) (Submit an original and a duplicate for fee processing)	5. <input type="checkbox"/> Microfiche Computer Program (Appendix)	
2. <input checked="" type="checkbox"/> Specification [Total Pages 12] (preferred arrangement set forth below) <ul style="list-style-type: none">- Descriptive title of the invention- Cross References to Related Applications- Statement Regarding Fed sponsored R & D- Reference to Microfiche Appendix- Background of the invention- Brief Summary of the invention- Brief Description of the Drawings (if filed)- Detailed Description- Claim(s)- Abstract of the Disclosure	6. Nucleotide and/or Amino Acid Sequence Submission (if applicable, all necessary) <ul style="list-style-type: none">a. <input type="checkbox"/> Computer Readable Copyb. <input type="checkbox"/> Paper Copy (identical to computer copy)c. <input type="checkbox"/> Statement verifying identity of above copies	
3. <input checked="" type="checkbox"/> Drawing(s) (35 U.S.C. 113) [Total Sheets 4]	ACCOMPANYING APPLICATION PARTS 7. <input checked="" type="checkbox"/> Assignment Papers (cover sheet & document(s)) 8. <input type="checkbox"/> 37 C.F.R. § 3.73(b) Statement (when there is an assignee) <input checked="" type="checkbox"/> Power of Attorney 9. <input type="checkbox"/> English Translation Document (if applicable) 10. <input type="checkbox"/> Information Disclosure Statement (IDS)/PTO-1449 <input type="checkbox"/> Copies of IDS Citations 11. <input type="checkbox"/> Preliminary Amendment 12. <input checked="" type="checkbox"/> Return Receipt Postcard (MPEP 503) (Should be specifically itemized) 13. <input type="checkbox"/> Small Entity Statement(s) <input type="checkbox"/> Statement filed in prior application, Status still proper and desired (PTO/SB/09-12) 14. <input type="checkbox"/> Certified Copy of Priority Document(s) (if foreign priority is claimed) 15. <input type="checkbox"/> Other: FILED CONCURRENTLY WITH 13DVI3046	
4. Oath or Declaration [Total Pages 2] <ul style="list-style-type: none">a. <input checked="" type="checkbox"/> Newly executed (original or copy)b. <input type="checkbox"/> Copy from a prior application (37 C.F.R. § 1.63(d)) (for continuation/divisional with Box 16 completed)<ul style="list-style-type: none">i. <input type="checkbox"/> DELETION OF INVENTOR(S) Signed statement attached deleting inventor(s) named in the prior application, see 37 C.F.R. §§ 1.63(d)(2) and 1.33(b).		
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METHODS AND APPARATUS FOR MODEL BASED DIAGNOSTICS

The Government has rights in this invention pursuant to Contract No. MDA972-98-3-0002 awarded by the Defense Advanced Research Projects Agency.

BACKGROUND OF THE INVENTION

The present invention relates generally to gas turbine engines, and more specifically, to diagnosing faults in such engines.

Gas turbine engines are used in aeronautical, marine, and industrial applications. In the aeronautical application, gas path or performance related faults are typically detected using flight-to-flight trending. Changes in sensed parameters are identified between a current flight and a previous flight. If multiple parameters are trended, then the pattern in the changes may be sufficiently distinct to allow classification (i.e., diagnosis) as a specific fault. With flight-to-flight trending, data scatter may occur, and such data scatter may be of a same order of magnitude as the fault effects to be identified.

BRIEF SUMMARY OF THE INVENTION

Systems and methods for performing model-based diagnostics are described. In an exemplary embodiment, sensor values from an actual engine plant are input to an engine component quality estimator which uses regression techniques to generate performance estimates of major rotating components, e.g. efficiency and airflow of a fan, compressor and turbines. Estimated performance differences are generating by comparing the generated performance estimates to a nominal quality engine. The estimated performance differences, which are indicative of component quality, are continuously updated and input to a real-time model of the engine. The model receives operating conditional data and the quality estimates are used to adjust the nominal values in the model to more closely match the model values to the actual plant. Outputs from the engine model are virtual parameters, such as stall margins, specific fuel consumption, and fan/compressor/turbine efficiencies. The virtual

parameters are combined with the sensor values from the actual engine plant in a fault detection and isolation classifier, such as a linear regressor or a neural network, to identify abnormal conditions and/or specific fault classes, and output a diagnosis.

BRIEF DESCRIPTION OF THE DRAWINGS

Figure 1 illustrates an absolute value - model based diagnostic system;

5 Figure 2 illustrates a physics based nonlinear model of a turbofan engine;

Figure 3 illustrates a neural network fault detection and isolation classifier;

10 Figure 4 illustrates a linear regressor fault detection and isolation classifier;

Figure 5 illustrates a sensor delta - model based diagnostics system;

Figure 6 illustrates a components quality delta - model based diagnostics system; and

15 Figure 7 illustrates a sensor delta-delta - model based diagnostics system.

DETAILED DESCRIPTION OF THE INVENTION

An exemplary embodiment of a system 10 for performing model-based diagnostics is illustrated in Figure 1. More specifically, and referring to Figure 1, sensor values 12 from an actual engine plant are input to an engine component quality estimator 14. Estimator 14 uses regression or other system identification techniques
20 to produce performance estimates of major rotating components, e.g. efficiency and airflow of a fan, compressor and turbines. Estimated performance differences are generating by comparing the generated performance estimates to a nominal quality engine. The estimated performance differences, which are indicative of component quality, are continuously updated and input to a real-time model 16 of the engine.

Model 16 receives operating conditional data 18 and the quality estimates are used to adjust the nominal values in model 16 to more closely match the model values to the actual plant. Outputs from engine model 16 are virtual parameters 20, such as stall margins, specific fuel consumption, and fan/compressor/turbine efficiencies. Virtual parameters 20 are combined with sensor values 12 from the actual engine plant in a fault detection and isolation classifier 22, such as a linear regressor or a neural network, to identify abnormal conditions and/or specific fault classes and output a diagnosis 24.

In a specific embodiment of engine component quality estimator 14, input vector x , representative of measured and modeled values of rotor speeds, temperatures, and pressures, is normalized and multiplied by a regressor matrix R , to obtain a vector of normalized component quality estimates \hat{y} (e.g., changes in efficiency and flow modifiers for the fan, booster, compressor, high pressure turbine, and low pressure turbine):

$$\hat{y} = \hat{x}R$$

where,

$$\hat{x} = (x - \bar{x}) / x_s$$

$$\hat{y} = (y - \bar{y}) / y_s$$

where,

\bar{x} and \bar{y} are mean values of x and y , respectively

x_s and y_s are scale factors for x and y , respectively

The regressor matrix R is generated by a 'training' process. In the training process, a large number of engines of specified and/or random component quality are simulated, and the resulting sensor values are obtained. In addition, a nominal engine or 'model' is run at the same operating conditions (altitude, mach number, total inlet temperature, power-setting parameter such as fan speed, and bleed setting, for example) as the engine simulation. The number of engines simulated is n , the number of component quality parameters is p , the number of engine sensor values

is a , and the number of nominal engine sensor or ‘model’ values is b . Then, a multiple linear regression solution provides the regressor matrix R using:

$$R = \hat{X} \setminus \hat{Y}$$

where \hat{X} and \hat{Y} are normalized values of X and Y obtained by the use of mean values and scale factors just like \hat{x} and \hat{y} above

and,

5 X is an $(a+b)$ by n matrix of sensor and model values

Y is a p by n matrix of simulated component quality variations

\setminus is the pseudo-inverse operator denoting a least-squares solution to $Y=XR$

10 In a typical implementation, data is collected at a single operating condition, or even a multiplicity of operating conditions or even operating regimes (such as takeoff, climb, and cruise). In another embodiment, estimator is a nonlinear estimator such as a neural network, rather than a linear regressor.

15 The linear regressor described above is a one-time or “snapshot” approach for estimating component health. For a continuous update, the regressor is replaced by a proportional plus integral (PI) regulator. Nominally, the proportional and integral gain matrices are set equal to the regressor matrix. However, these matrices can be modified, for instance by multiplying by a constant that can be tuned, to modify regulator performance.

20 Model 16 of the plant, or engine, is used to estimate sensed parameters such as rotor speeds, temperatures, and pressures, given environmental conditions, power setting parameters, and actuator positions as input. Model 16 is, for example, a physics-based model, a regression fit, or a neural-net model of the engine. In an exemplary embodiment, model 16 is a physics-based aerothermodynamic model of the engine. This type model is referred to as a Component Level Model (CLM) because
25 each major component in the engine (e.g., fan, compressor, combustor, turbines,

ducts, and nozzle) is individually modeled, and then the components are assembled into the CLM.

Figure 2 illustrates components of engine model 16. As shown in Figure 2, model 16 includes an air inlet 50 and a fan 52 downstream from inlet 50. Model 16 also includes, in series flow relationship, a booster 54, a compressor 56, a burner 58, and a high pressure turbine 60, and a low pressure turbine 62. Exhaust flows from a core nozzle 64, which is downstream from low pressure turbine 62. Air also is supplied from fan 52 to a bypass duct 66 and to a bypass nozzle 68. Exhaust flows from bypass nozzle 68.

The CLM is a fast running transient engine cycle representation, with realistic sensitivities to flight conditions, control variable inputs and high-pressure compressor bleed. The CLM is tuned to match actual engine test data both for steady-state and transient operation.

A specific embodiment of fault detection and isolation classifier 22 is illustrated in Figure 3. Engine sensor values 12 and model parameters 20 are input to a feed-forward neural network 80. Outputs from neural network 80 include indicators for fault paths 82 appropriate to each engine rotating component as well as a no-fault path 84. Network 80 is trained on large sets of engine/model data that includes effects of engine quality, deterioration, sensor bias, and operating conditions. These data sets include predominantly unfaulted data as well as representative sets of engine/model data for each fault to be classified.

An alternate embodiment of fault detection and isolation classifier 22 is illustrated in Figure 4. Engine sensor values 12 and model parameters 20 are input to a linear regressor 90. Regressor 90 is similar to the regressor described above, except that the 'training' data includes simulated unfaulted engines as well as simulated engines with specific faults of varying magnitudes.

System 10 described above provides the advantage that it does not rely on prior knowledge or collection of data from a prior operating condition. System 10 therefore is suited for diagnosing pre-existing faults, i.e. faults that are present when

the engine is initially started. An example of a pre-existing fault is a variable stator vane misrigging.

Figure 5 illustrates a sensor delta - model based diagnostics system 100. Components / processing in system 100 identical to components / processing in system 10 are identified in Figure 5 using the same reference numerals as used in Figure 1. Referring now specifically to Figure 5, sensor values 12 from the actual engine plant are input to component quality estimator 14. Estimator uses regression or system identification techniques to produce performance estimates of the major rotating components, e.g. efficiency and airflow of the fan, compressor and turbines, which are relative to a nominal quality engine. These estimated performance differences, which are indicative of the quality of these components, are input one time, typically at the beginning of a flight to a real-time model 16 of the engine.

Model 16 receives the same operating conditional data 18 as the actual engine plant, and the quality estimates are used to adjust the nominal values in model 16 in order to more closely match those of the actual plant. It is assumed at this time that the actual engine plant is unfaulted. Outputs from engine model 16 are estimates of the engine sensors 102. Computed sensor values 102 are deducted 104 from the actual engine sensor values 12 to produce sensor deltas 106. Deltas 106 should normally be close to zero for a no-fault condition. When a fault occurs the actual engine sensor values 12 will change, but the model computed sensors 102 will not. As a result, the values of the sensor deltas 106 will change. Sensor deltas 106, together with the engine operating conditions 18, are then input to fault detection and isolation classifier 22 such as a neural network or linear regressor, to identify abnormal conditions and/or specific fault classes and output a diagnosis 108.

System 100 provides the advantage that model 16 is adjusted to account for sensor bias effects and engine component quality. An accurate quality estimation from estimator 14 is needed in order to determine the sensor deltas 106.

Figure 6 illustrates an estimated quality deltas model-based diagnostic system 200. Components / processing in system 200 which are identical to

components / processing in system 10 are identified in Figure 6 using the same reference numerals as used in Figure 1. Referring now specifically to Figure 6, sensor values 12 from the actual engine plant together with estimated sensor values 20 from model 16 of a nominal (new, unfaulted) engine are input to a component quality estimator 14. Sensor and model values from a multiplicity of operating condition are used. In this embodiment, data 18 from two engine operating conditions, e.g., pre-flight ground idle and takeoff, are utilized in component quality estimator 14 to produce an initial (pre-flight) quality estimate 202 of the major rotating components, e.g. efficiency and airflow of the fan, compressor and turbines. This process is repeated at the end of the flight by using inflight data such as the last high power data set (before descent) and post-flight ground idle 18 from the engine sensors 12 and the associated model computed sensor data 20. The resulting output from the component quality estimator 14 produces a final (post-flight) quality estimate 204. Component quality estimator 14 is the same estimator as used pre-flight except the gains used in post-flight are modified due to the difference in operating conditions at which the data is collected.

A difference 206 between final quality estimate 204 and initial quality estimate 202 is determined. A delta change in quality 208 is then be assessed in fault detection and isolation classifier 22, such as a neural network or linear regressor, to identify abnormal conditions and/or specific fault classes and output a diagnosis 210.

System 200 directly identifies engine quality changes, which facilitates significantly reducing the complexity of diagnostic classification assessment. The absolute values of the individual quality estimates need not be accurate, only consistent, since the resulting delta quality (i.e., quality value 208) is indicative of the fault effects.

Figure 7 illustrates a sensor delta-delta model-based diagnostic system 300. Components / processing in system 300 which are identical to components / processing in system 10 are identified in Figure 7 using the same reference numerals as used in Figure 1. Referring now specifically to Figure 7, a real-time engine model 16 is run to the same operating conditions 18 as the actual engine plant. Model 16

represents a nominal unfaulted engine, or optionally it could be updated at the beginning of a flight with quality adjustments 302 from a component quality estimator. The outputs from real-time engine model 16 are estimates of engine sensors 304. These estimates are compared (e.g., subtracted) 306 with actual sensor values 12 from the engine plant and deltas (e.g., delta 1 and delta 2) are determined. The engine plant is assumed to be unfaulted and the deltas can therefore be considered as a reference (normal) set 308.

During flight, as engine operating conditions 18 change, new estimated sensor values 310 are computed by real-time engine model 16. The associated actual engine sensor values 12 can then be compared 312 with estimates 310 to compute new (current) deltas 314. If the engine is still unfaulted, current deltas 314 will be similar to initial reference deltas 308 and a change 316 in deltas, or delta-deltas 318, will be close to zero. When a fault occurs current deltas 314, a change relative to the reference deltas 316 produces non-zero delta-deltas 318. Delta-deltas 318 can then be input together with engine operating conditions 18 to fault detection and isolation classifier 22, such as a neural network or linear regressor, to identify abnormal conditions and/or specific fault classes and output a diagnosis 320. Engine operating conditions 18 are utilized in the classifier since the effect of engine faults may not constant throughout the flight envelope.

System 300 improves upon system 100 in that in system 300, residual effects of sensor bias and component quality that are not accounted for by the model adjustments in system 100 are negated by the delta-delta determination in system 300.

While the invention has been described in terms of various specific embodiments, those skilled in the art will recognize that the invention can be practiced with modification within the spirit and scope of the claims.

WHAT IS CLAIMED IS:

1. A system for monitoring engine performance, said system comprising:

an engine model;

a component quality estimator coupled to said engine model; and

5 a fault detection and isolation classifier coupled to said engine model and to said component quality estimator.

2. A system in accordance with Claim 1 wherein said model comprises a component level model of an engine.

3. A system in accordance with Claim 1 wherein said component
10 quality estimator comprises a linear regressor.

4. A system in accordance with Claim 1 wherein said fault detection and isolation classifier comprises a feed-forward neural network.

5. A system in accordance with Claim 1 wherein said fault detection and isolation classifier comprises a linear regressor.

6. A system in accordance with Claim 1 wherein said engine
15 component quality estimator generates performance estimates, and said model generates virtual parameters based on the performance estimates and engine operating conditions.

7. A system in accordance with Claim 6 wherein said fault
20 detection and isolation classifier identifies fault conditions based on said virtual parameters and engine sensor values.

8. A system in accordance with Claim 6 wherein said engine virtual parameters are subtracted from engine sensor values to generate sensor deltas supplied to said fault detection and isolation classifier.

9. A system in accordance with Claim 8 wherein reference sensor deltas are generated and are compared to later obtained sensor deltas to generate sensor delta-deltas supplied to said fault detection and isolation classifier.

10. A system in accordance with Claim 1 wherein said model generates virtual sensor values based on engine operating conditions, and said component quality estimator generates quality estimates based on said virtual sensor values and engine sensor values.

11. A system in accordance with Claim 10 wherein said engine component quality estimator generates an initial quality estimate and a final quality estimates, and the initial and final quality estimates are compared to generate a delta quality supplied to said fault detection and isolation classifier.

12. A method for monitoring engine performance, said method comprising the steps of:

supplying engine operating conditions to an engine model;

supplying engine sensor values to an engine component quality estimator;

generating a fault detection based at least in part on engine model outputs, and at least in part on estimator outputs.

13. A method in accordance with Claim 12 further comprising the step of supplying the estimator outputs to the engine model.

14. A method in accordance with Claim 13 wherein the engine model generates sensor estimates, and said method further comprises the step of generating sensor deltas by comparing the sensor estimates to engine sensor values.

15. A method in accordance with Claim 12 further comprising the step of supplying the engine model outputs to the engine component quality estimator.

16. A method in accordance with Claim 15 further comprising the steps of:

operating the engine component quality estimator to generate an initial quality estimate and a final quality estimate; and

comparing the initial quality estimate to the final quality estimate to generate a delta quality.

5 17. A method in accordance with Claim 13 wherein the engine model generates sensor estimates, and said method further comprises the steps of:

generating a reference sensor delta by comparing the sensor estimates at a first operating point;

generating a current sensor delta at a second operating point; and

10 generating a sensor delta-delta by comparing the reference sensor delta to the current sensor delta.

18. A method for monitoring engine performance, said method comprising the steps of:

15 generating a reference sensor delta by comparing sensor values with sensor estimates at a first operating point;

generating a current sensor delta at a second operating point;

generating a sensor delta-delta by comparing the reference sensor delta to the current sensor delta; and

generating a fault detection based upon the sensor delta-deltas.

METHODS AND APPARATUS FOR MODEL BASED DIAGNOSTICS

ABSTRACT OF THE DISCLOSURE

Systems and methods for performing module-based diagnostics are described. In an exemplary embodiment, sensor values from an actual engine plant are input to an engine component quality estimator which generates performance estimates of major rotating components. Estimated performance differences are generating by comparing the generated performance estimates to a nominal quality engine. The estimated performance differences, which are indicative of component quality, are continuously updated and input to a real-time model of the engine. The model receives operating conditional data and the quality estimates are used to adjust the nominal values in the model to more closely match the model values to the actual plant. Outputs from the engine model are virtual parameters, such as stall margins, specific fuel consumption, and fan/compressor/turbine efficiencies. The virtual parameters are combined with the sensor values from the actual engine plant in a fault detection and isolation classifier to identify abnormal conditions and/or specific fault classes, and output a diagnosis.

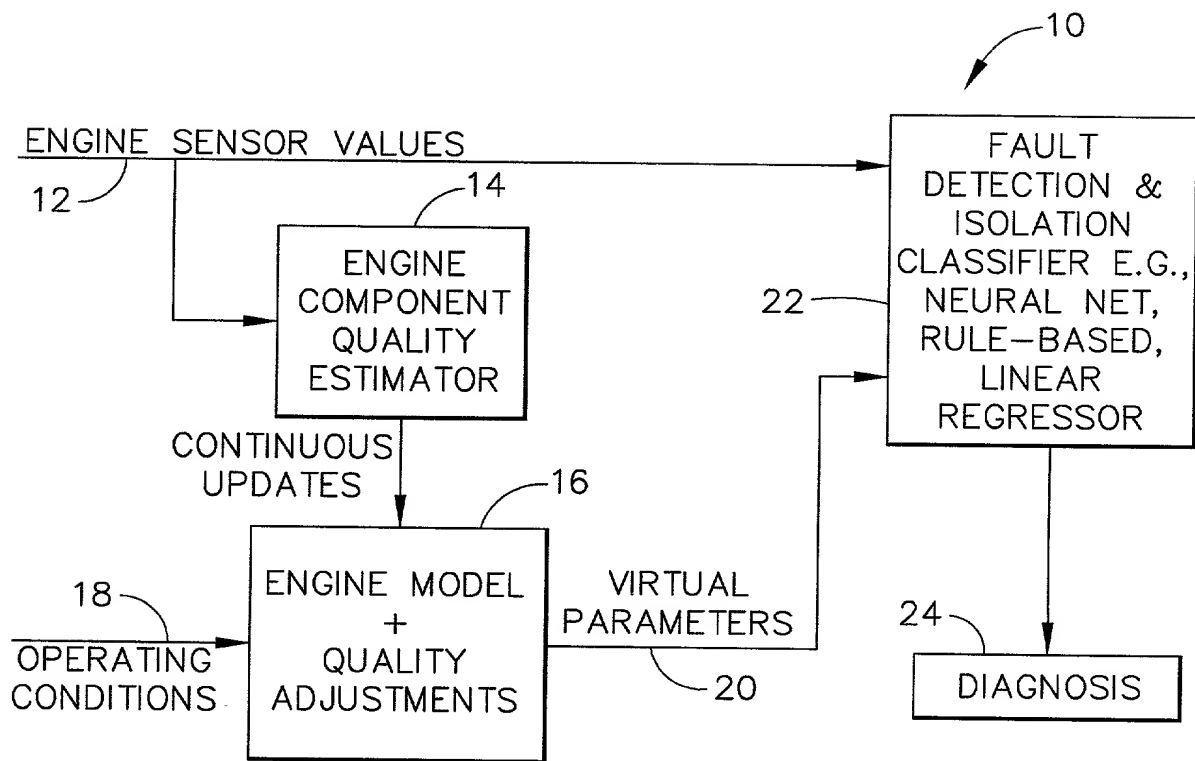


FIG. 1

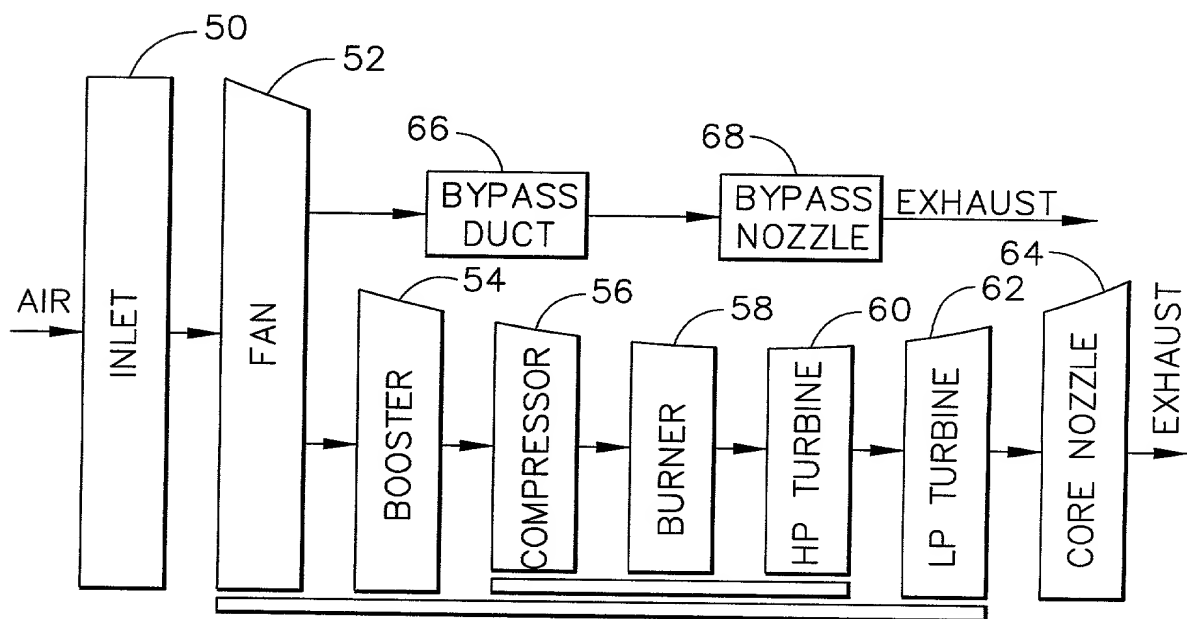


FIG. 2

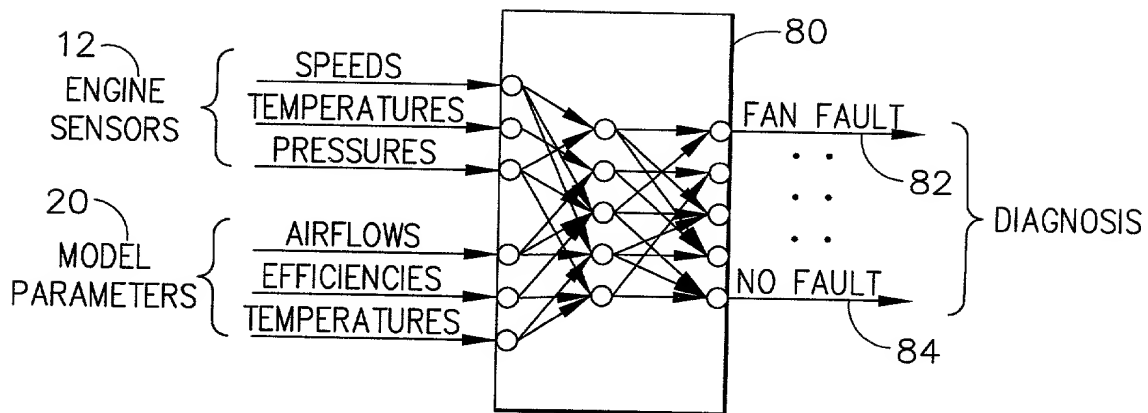


FIG. 3

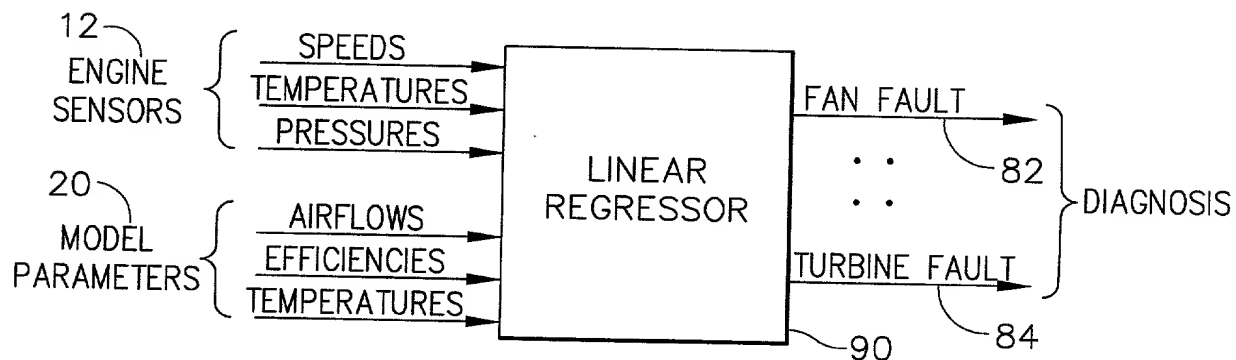


FIG. 4

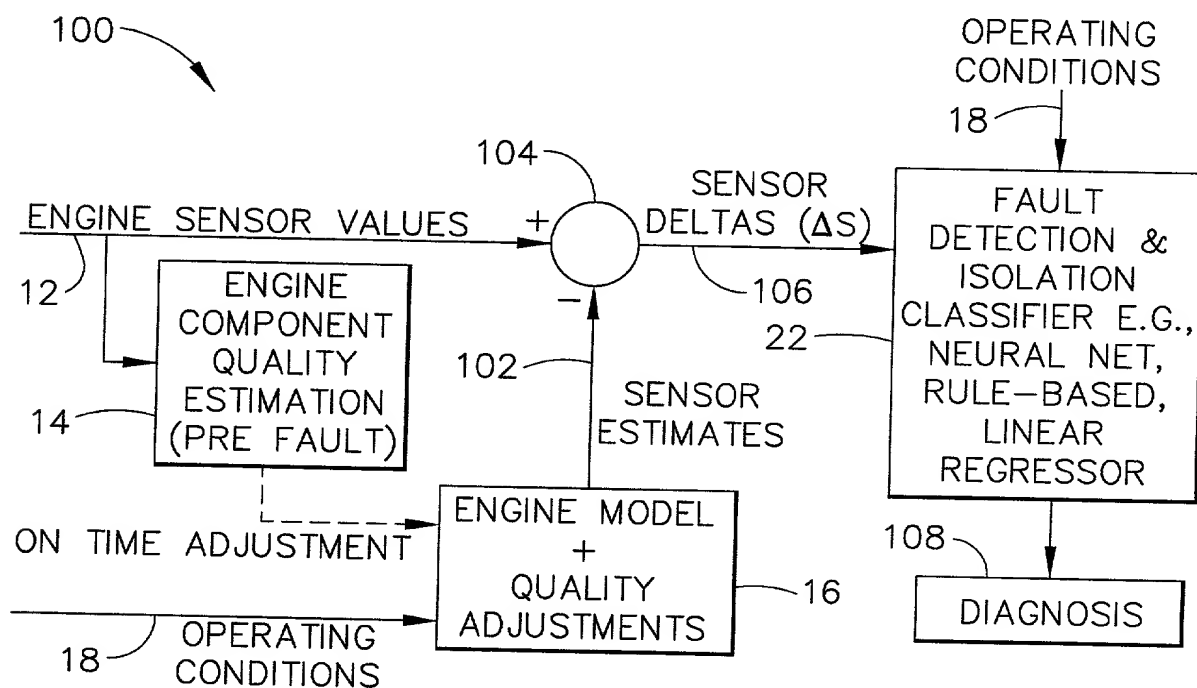


FIG. 5

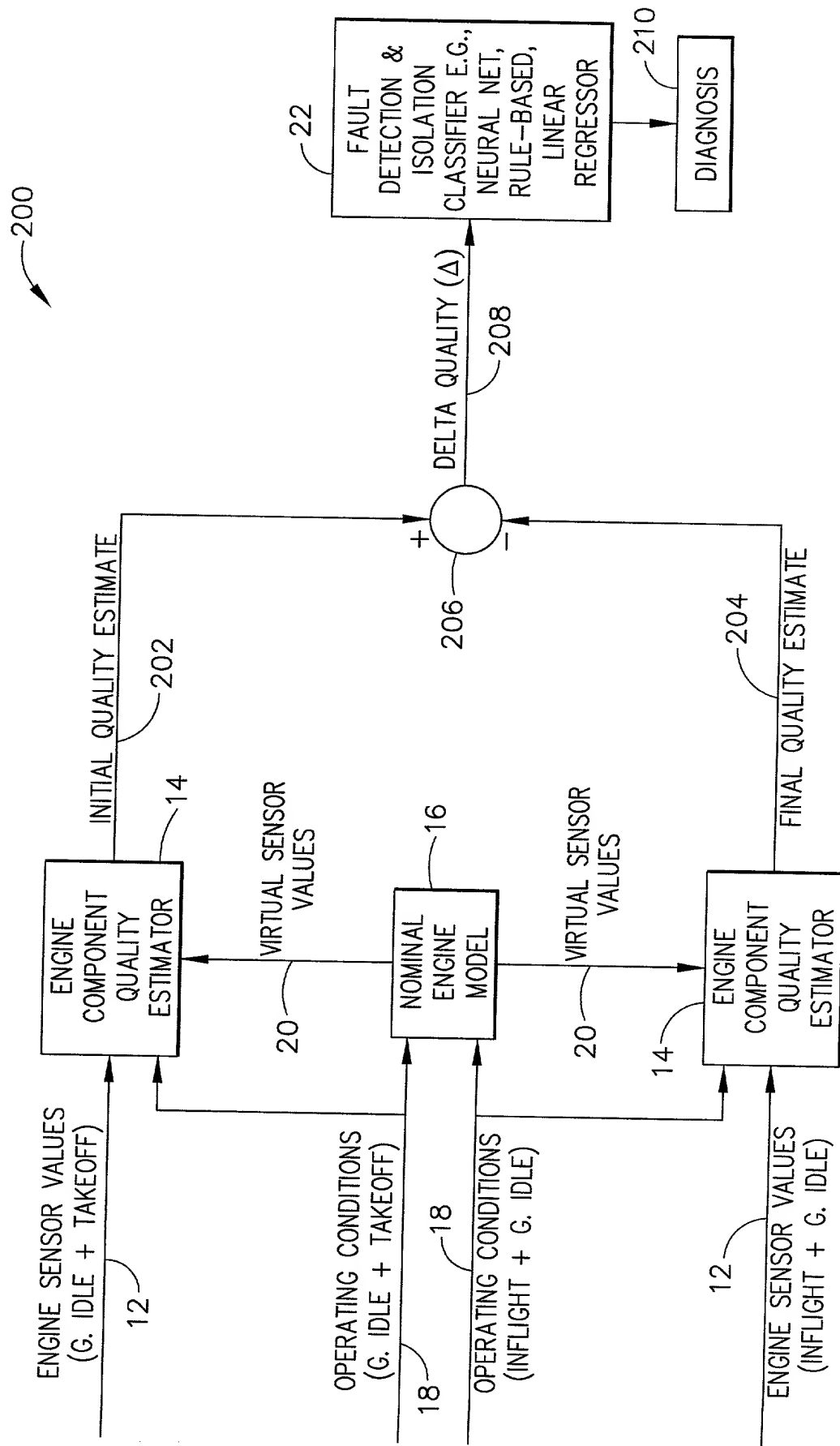


FIG. 6

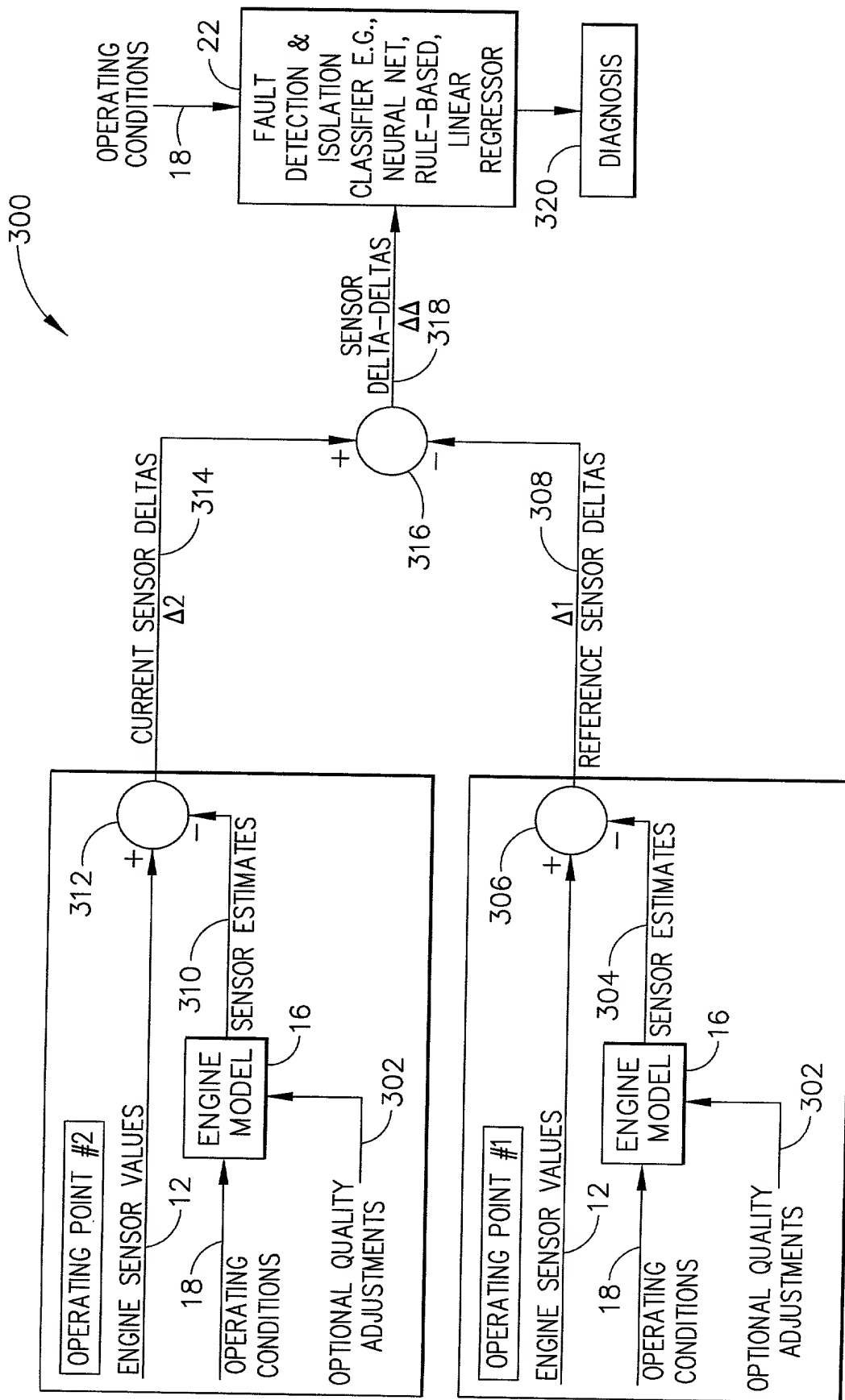


FIG. 7

**DECLARATION AND POWER OF ATTORNEY
FOR PATENT APPLICATION**

Docket Number
13DV13092

As a below named inventor, I hereby declare that:

My residence, post office address, and citizenship are as stated below next to my name.

I believe I am the original, first and sole inventor (if only one name is listed below) or an original, first and joint inventor (if plural names are listed below) of the subject matter which is claimed and for which a patent is sought on the invention entitled:

METHODS AND APPARATUS FOR MODEL BASED DIAGNOSTICS

the specification of which

☒ is attached hereto
OR

☐ was filed on _____ as United States Application Number or PCT International Application Number _____
and was amended on _____ (if applicable).

I hereby state that I have reviewed and understand the contents of the above identified specification, including the claims, as amended by any amendment specifically referred to above.

I acknowledge the duty to disclose information which is material to patentability as defined in Title 37 Code of Federal Regulations, §1.56.

I hereby claim foreign priority benefits under Title 35, United States Code §119 (a)-(d) or §365 (b) of any foreign application(s) for patent or inventor's certificate, or §365 (a) of any PCT international application which designated at least one country other than the United States of America, listed below and have also identified below any foreign application for patent or inventor's certificate, or of any PCT international application having a filing date before that of the application on which priority is claimed.

PRIOR FOREIGN APPLICATION(s)

Priority Claimed

(Number)

(Country)

(Day/Month/Year Filed)

☐ Yes ☐ No

(Number)

(Country)

(Day/Month/Year Filed)

☐ Yes ☐ No

☐ Additional foreign application numbers are listed on a supplemental priority data sheet attached hereto.

I hereby claim the benefit under Title 35, United States Code §119 (e) of any United States provisional application(s) listed below.

(Application Number)

(Filing Date)

☐ Additional provisional application numbers are listed on a supplemental priority data sheet attached hereto.

I hereby claim the benefit under Title 35, United States Code §120 of any United States Application(s), or §365 (c) of any PCT international application designating the United States of America, listed below and, insofar as the subject matter of each of the claims of this application is not disclosed in the prior United States or PCT International application in the manner provided by the first paragraph of Title 35, United States Code §112, I acknowledge the duty to disclose information which is material to patentability as defined in Title 37, Code of Federal Regulations §1.56 which became available between the filing date of the prior application and the national or PCT international filing date of this application.

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(Filing Date)

(Status - patented, pending, abandoned)

(Application Number)

(Filing Date)

(Status - patented, pending, abandoned)

I hereby appoint the registered practitioners associated with Customer Number 006111 to prosecute this application and to transact all business in the Patent and Trademark Office connected therewith.

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Last Name

Signature: _____

Date

Residence: _____

City and State

Citizenship: _____

Post Office Address: _____

FOURTH JOINT INVENTOR:

Full name: _____

First Name

Middle Name

Last Name

Signature: _____

Date

Residence: _____

City and State

Citizenship: _____

Post Office Address: _____